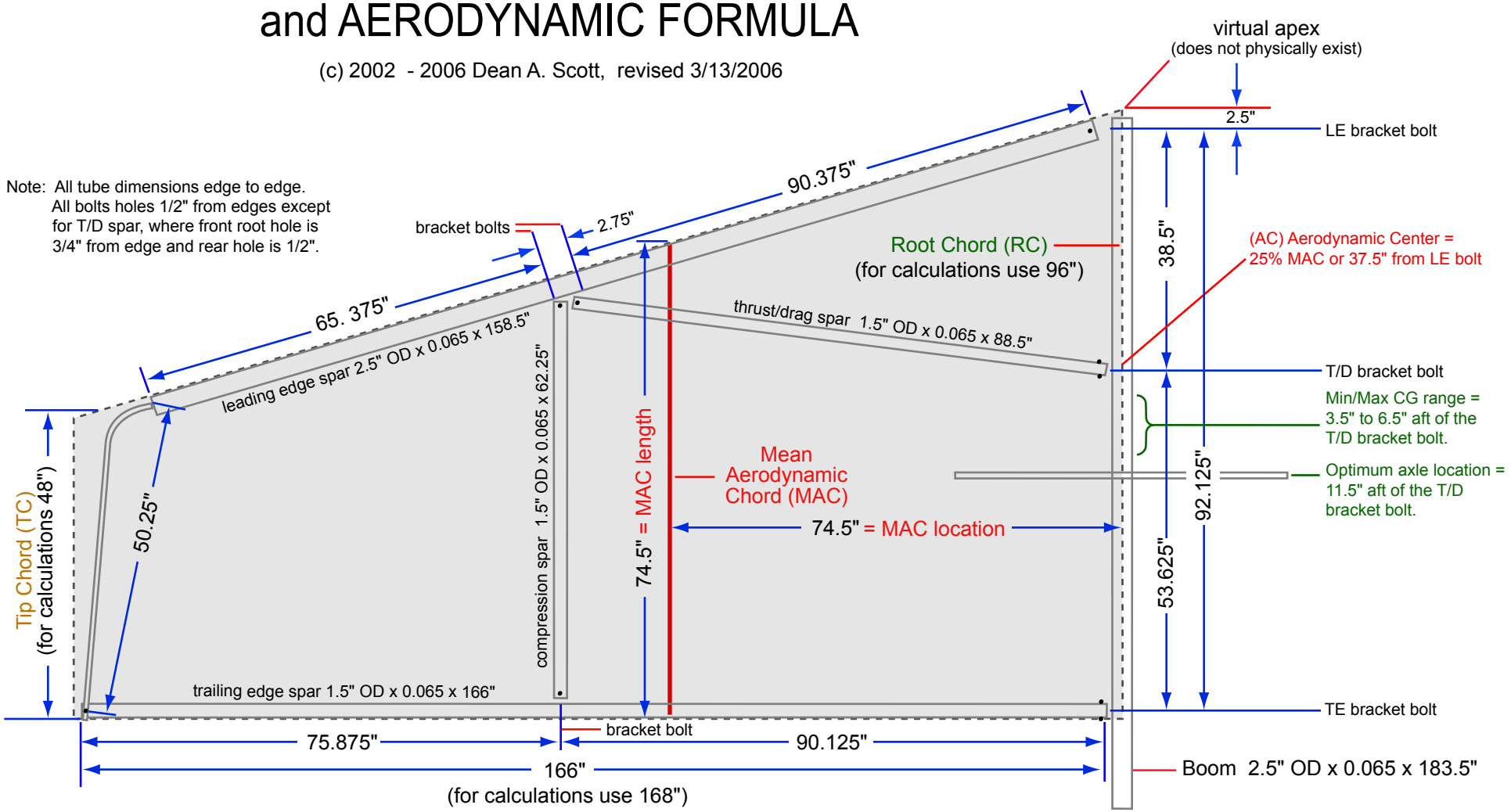
WEEDHOPPER[tm] MODEL C, MODEL 40, SUPERSINGLE, and TWO-PLACE WING GEOMETRY and AERODYNAMIC FORMULA



MEAN AERODYNAMIC CHORD (MAC) *

$$\frac{2 * (1 + (TC / RC) + (TC / RC)^2)) * RC}{3 * (1 + (TC / RC))}$$

MAC SPAN LOCATION

(0.5 * Wing Span) * (RC - MAC) / (RC - TC)

CG as %MAC

(1 - ((RC - (CG + 2.5)) / MAC)) * 100

where CG = inches aft of LE bolt

CENTER OF LIFT (COL) level flight *

RC - (MAC * 0.7)

AERODYNAMIC CENTER (AC) fixed *

RC - (MAC * 0.75)

* subtract 2.5" from calculation results to use LE bracket bolt as the measurement reference